



NASA and Cryogenic Technology Applications

Liquid Hydrogen Technologies Workshop

hosted by

The U.S. Department of Energy's Hydrogen and Fuel Cell Technologies Office and
the National Aeronautics and Space Administration's Cryogenic Technical Discipline Team

February 22-23, 2022

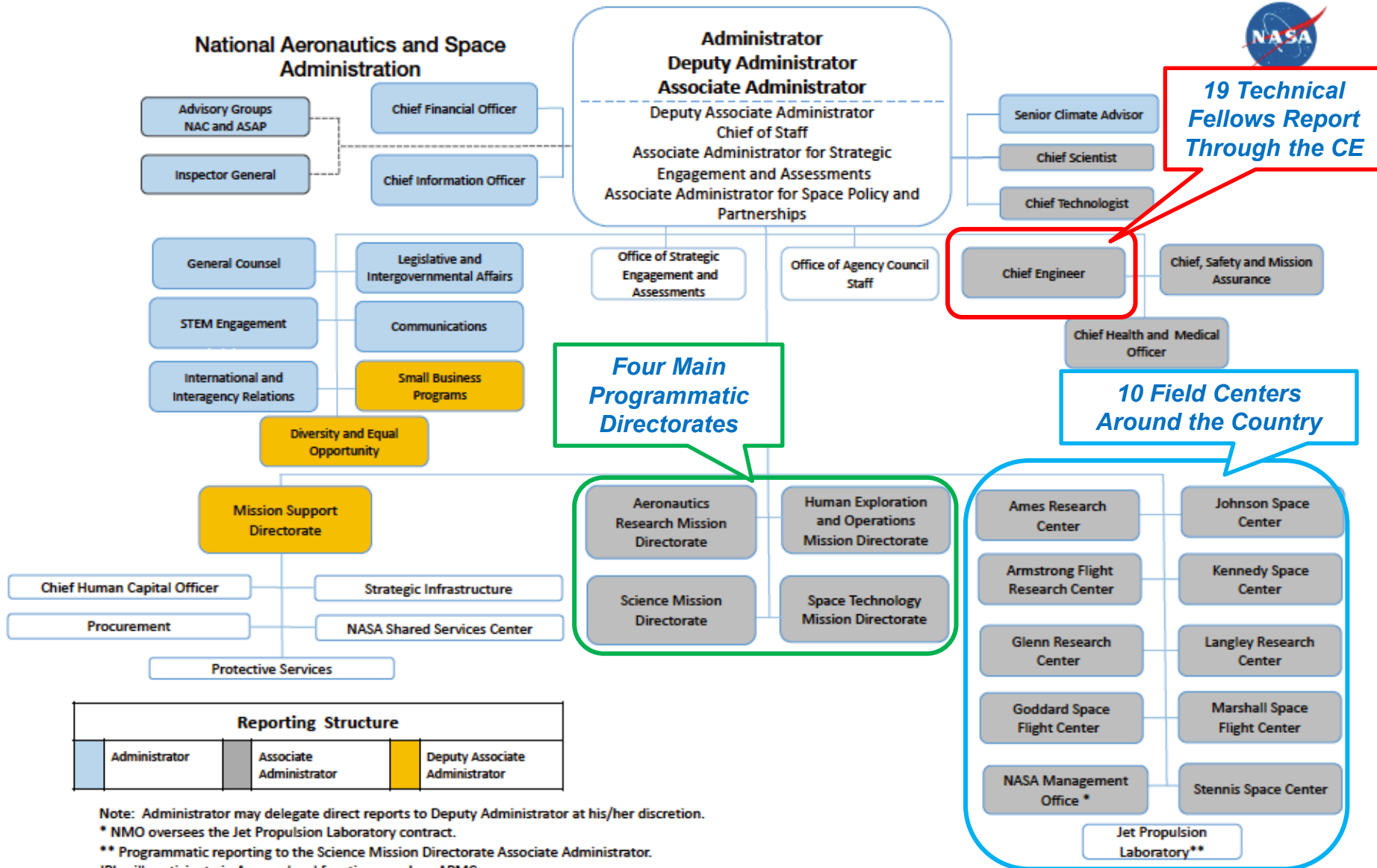
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NASA Organizational Structure



Decomposition of Cryogenics at NASA

- Scope of the Cryogenics Technical Discipline Team (TDT)



Thermal conditioning for Sensors, Instruments, and High Efficiency Electronic Motors

- Thermal parasitics
- Refrigeration below 10K
- Refrigeration above 10K
- Solid cryogenics heat sink
- Solar Shields
- Coatings
- Heat switches

In-space Propellant Storage & Utilization

- Vacuum/partial vacuum insulation
- Micro-g fluid dynamics
- PMDs
- Gauging
- Pressure control
- CFD
- Cryocooling/zero boil-off
- Propellant Transfer
- Liquefaction

Launch Vehicle Propellant

- Atmospheric tank/line insulation
- Stratification
- Slosh/ullage collapse
- Feedline chill
- Geysering
- Mass Gauging
- Quick Disconnects

Ground Testing and Operations

- Atmospheric Insulation
- Densification
- Large Scale Refrigeration
- Quick Disconnects
- Cryogenic Pumps
- Leak/Fire Detection
- Automation/Fault Detection

Continued expertise in Cryogenic Analysis, Safety, & Properties are key to success in all areas of the discipline

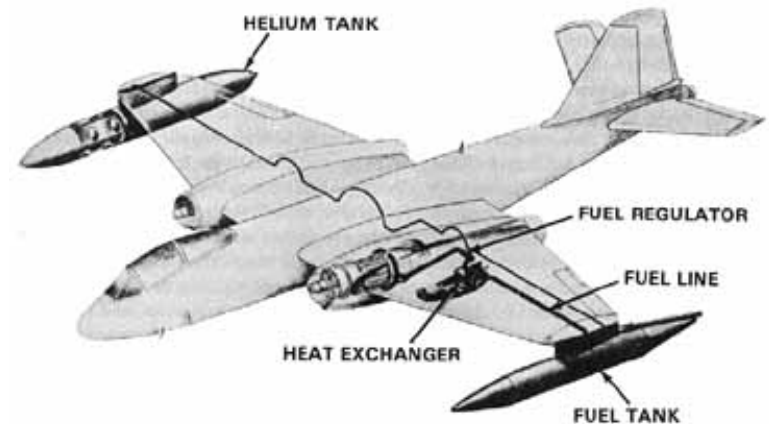
“Aeronautics” Flew with Liquid Hydrogen Before NASA Was Created



Then

NACA Lewis (now Glenn) conducts Project Bee (1955-1959)

- B-57B modified to permit one engine to burn JP-4 or H₂



Now

The Center for High-Efficiency Electrical Technologies for Aircraft (CHEETA)

- A multi-disciplinary consortium of researchers, scientists, and engineers from a variety of universities, laboratories, and industry groups.
- Studying electric aviation including superconductive components cooled by LH2 with H2 fuel cell generated electricity.



CHEETA

Center for High-Efficiency Electrical
Technologies for Aircraft



Launch Systems and Ground Testing Systems

Space Launch System

- Thrust = 8.8-11.9 Mlbs
- Total On-Board Cryo Prop.
- $\text{LO}_2 = 995 \text{ m}^3$, $\text{LH}_2 = 1,770 \text{ m}^3$
- First Launch: Spring 2022

98 m

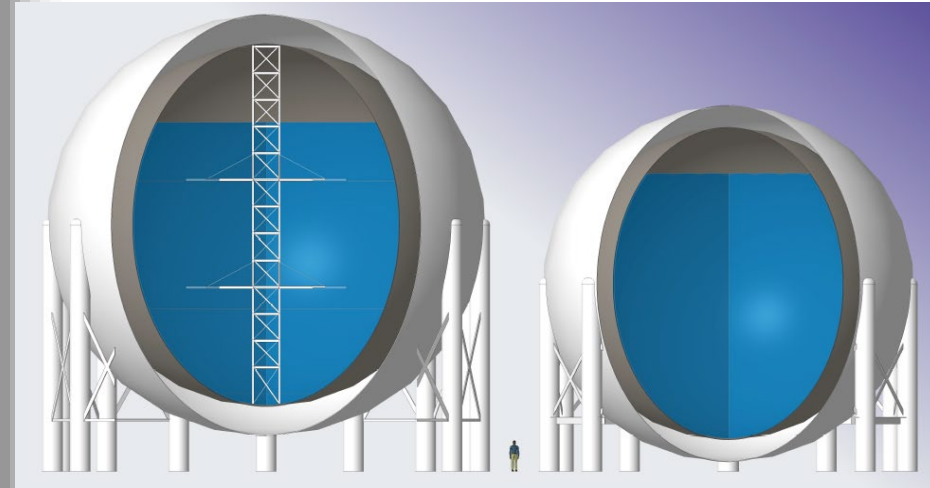


New LH2 Sphere

- 4,733 m^3
- Glass bubble insulation
- Integrated refrigeration HX

Apollo LH2 Sphere

- 3400 m^3



Integrated Refrigeration Testing

- 125 m^3 LH2



RS-25 Engine Testing

- 162 lb/sec LH2
- 970 lb/sec LO2

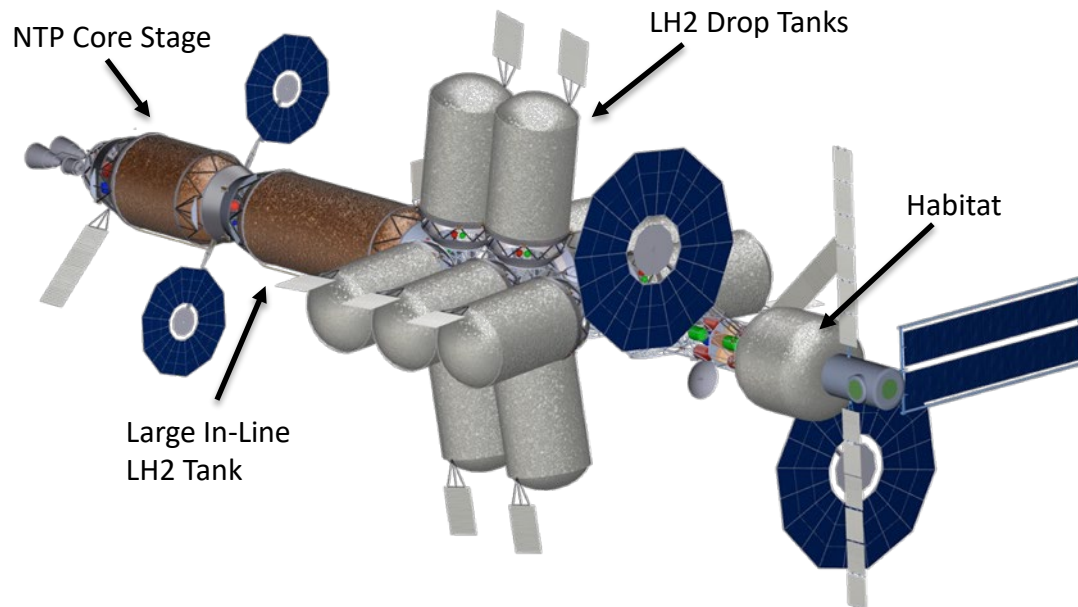


What Might a Crewed Mars Mission Look Like?



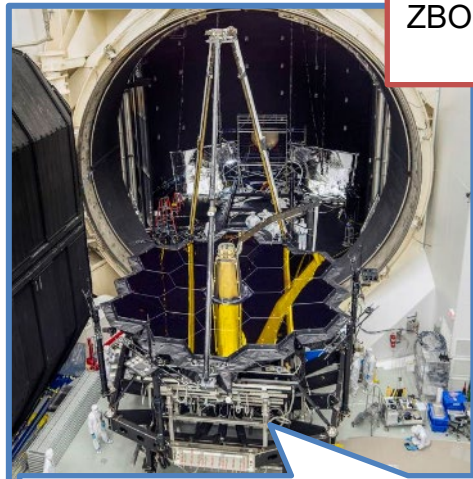
The orbital mechanics generally require a 2 – 3 year round trip and requires a very large amount of propulsive energy

- One concept uses a Nuclear Thermal Propulsion (NTP)
 - Reactor, LH2 pump, hydrogen heat exchange, and a converging expanding nozzle to generate thrust
- Liquid hydrogen storage in multiple large tanks (high performance passive thermal control with integrated refrigeration for zero LH2 loss)
- Large habitat for crew
- In-space assembly



Strategic/Key Facilities and Assets

Key cryogenic facilities encompass a wide range of sizes, types, and capabilities



JWST emerging from Chamber A (**JSC**)

ZBO Test Article at the Small Multi-Purpose Research Facility (SMiRF) (**GRC**)



Test Sample being removed at the Hydrogen Test Facility (HTF) (**MSFC**)



NGAS HEC Cryocooler in the CSE 3-ft Vacuum Chamber (**JPL**)



ISCCPD Cryogenic Component Test Facility (**ARC**)



Cryogenics Test Lab (**KSC**)



240,000 Gallon Liquid Hydrogen Supply Barges (**SSC**)



Cryogenic Research and Integration Facility (CRIF) (**GSFC**)